

Introduction

The Equator-S mission

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Abstract. In spite of its short operational life of only five months, the major goals of the Equator-S mission were fulfilled, except that its contribution to the ISTP science is restricted to the morning sector of the outer magnetosphere. A set of twelve papers following this introduction is a first documentation of the achievements. They span from the successful testing and operation of the most advanced and complex way of measuring electric fields in a hot plasma environment by means of electron beams, to various investigations at or near the equatorial magnetopause and in the plasma sheet.

1 Introduction

The Equator-S mission was different from most contemporary scientific satellite missions in that spacecraft and mission were designed and developed at a research institute, the Max-Planck-Institut für extraterrestrische Physik (MPE) with a few items supplied by agencies and industry. Only for mission analysis and operations the German Space Operations Center (GSOC) stepped in. Neither the European nor the German space agency was in the position to accommodate this mission in their regular programs within the desired time frame. Only a low-cost approach, enabled by the past satellite-building experience of MPE, offered a solution and was accepted for funding by the then still existing German Space Agency (DARA) in December 1994. Support promised and subsequently given by ESA and NASA was important for that decision. This paper will briefly review the motivation for this mission, the spacecraft development, the short mission history, and finally introduce the subsequent set of twelve papers presenting first results and discoveries.

2 Motivation

The primary objective of MPE emerged from its responsibility for the Electron Drift Instrument (EDI)

on the Cluster mission. EDI uses an electron beam technique to determine the electric field by sensing the displacement of electrons during one gyroperiod. In order to cope with any possible direction of the magnetic field and with its rapid changes, electron gun and detector were designed to allow beam injection and detection into and from any direction within a half sphere. The operation of EDI, i.e., the onboard program for the tracking algorithm, is exceedingly complex. Hence, the EDI team had a high interest in learning to employ this method in space before the launch of Cluster, because it was feared that the learning curve may be rather steep, leading to substantial data losses when first attempted on the four-satellite Cluster mission.

Hence, one was looking for a test flight opportunity. Since return of the beam to the detector is only possible if it is emitted into the plane transverse to the magnetic field with an accuracy better than 0.5°, instantaneous measurements of the field have to be included into the tracking algorithm, and thus the payload had to include a very sensitive and stable magnetometer. The chance of a realization appeared in late 1990, when the general idea of the Equator-S mission was born. But, of course, one would not design a satellite mission just with EDI and a magnetometer. The final mission concept was to contribute substantially to the ISTP program by choosing an eccentric equatorial orbit and thus filling the equatorial gap of that program (hence the name Equator-S, with *S* for small) and by adding a few key particle instruments. As a third element, all measurements, including those of 3D particle distributions, were to be done with the highest possible temporal resolution. To this end, a spin period of 1.5 sec was chosen, which had strong implications on the S/C design.

The search for economical launch opportunities took several years, until by the end of 1994, a launch as Auxiliary Passenger on an Ariane 4 appeared affordable to DARA, and green light was given for this mission. Although, the hope to launch Equator-S before Cluster had meanwhile faded away, the mission was still judged

to be worthwhile because of its contribution to the ISTP. When the launch of Cluster failed in June 1996 and ESA decided to have a recovery Cluster-II mission with a launch in mid-2000, the original primary goal was re-established.

3 Spacecraft development

A group of experimenters had gathered around the Equator-S planning team already in early 1991. The core group consisted of participants in the Equator mission of NASA's Global Geospace Science program, a mission cancelled in 1986. But NASA had promised to support these teams, in case an opportunity for replacement showed up. When, during the subsequent years, it became a realistic possibility to build a reasonably sized spacecraft, the payload summarized in Table 1 was accepted and an appropriate spacecraft designed. Furthermore, two technological experiments (GPS and solar cell diagnostics) were added to the primary payload. Designs and even some instrument hardware were taken from the Cluster and Wind missions. The table also lists the key institutes and some of the actors involved.

The design criteria of the spacecraft were as follows. It had to be built around a solid fuel kick-motor (provided by NASA) needed for injection from geostationary transfer orbit into the final orbit. Various particle detectors and the electron gun-detector units had to be accommodated with their particular viewing conditions. Two magnetometers each were mounted on two rigid 1.8-m booms. The spacecraft surface had to provide space for solar panels providing an average power of at least 60 W. For attitude control and spin-up, magnetic torquing was chosen. This implied the installation of circular and linear coils. They had to be operated in the stronger geomagnetic field near perigee, in order to finally achieve stable isolation by rotating the spin-axis out of the orbital plane to perpendicular to the ecliptic plane, a procedure requiring several months. The spacecraft had to fit into the Ariane 4 passenger capsule and needed altogether five TM antennas in order to cope with all potential visibility constraints from the ground station at Weilheim. A detailed description of the spacecraft was given by Stöcker *et al.* (1996).

For the electrical system, cost reasons required that almost exclusively left-over parts of the AMPTE and

Rosat missions were used. The flight system unit was shielded with 8 mm Al against the harsh radiation environment in the eccentric equatorial orbit. The solar panels had highly efficient GaAs cells and were provided and mounted by German industry. In order to achieve a data transmission with minimal contact times with the ground station, a mass memory with 192 MBytes was provided by UNH and funded by NASA.

The spacecraft was developed and built at MPE between January 1995 and September 1997 by a team of about eight persons with involvement of the workshops and other facilities of the institute and of industry. The launch mass was 229 kg. The costs of the spacecraft, including the MPE effort, were approximately 8.5 MioDM. Figure 1 shows the spacecraft with booms not extended. Two major instruments, EDI and ICI (see Table 1), and several antennas on the bottom side are well visible.

4 Mission history

Equator-S was launched on 2 December 1997 from French Guyana as auxiliary passenger on an Ariane 4 together with a Japanese telecom satellite. After spin-up to 48 rpm with help of the torquing system, it was injected into the final orbit by the kick-motor on 11 December 1997. The orbital characteristics are: 497 km perigee, 67275 km height of apogee or 11.5 R_E geocentric distance, 3.9 degrees inclination, and 22.3-hour period. The commissioning phase lasted into February 1998.

Unfortunately, the mission was damaged by several malfunctions. On 17 December 1997 the primary master processor unit failed, and on 30 April 1998 the second unit experienced the same problem. This caused the premature end of the mission. Failure analysis attributed both events to a latch-up in one the RAM-chips used by the processors. It is not unlikely that deep dielectric charging by the high energetic electron fluxes in December 1997 as well as in April 1998 was the cause of the latch-ups. Unfortunately, that part of the processor system had no latch-up protection and for safety reasons no switch-off capability was installed.

Besides the rather short mission duration (a minimum of 15 months was planned) a serious instrument failure occurred. In January 1998, the 3D Electron Analyzer suffered a sudden failure of all pre-amplifiers, which was caused by an undetected software error. As a

Table 1. Scientific payload

Name	Instrument type	Measurement	Lead Investigator	Institution
MAM	Fluxgate magnetometer	Magnetic field, 0–64 Hz	W. Baumjohann	MPE Garching
EDI	Electron beam sensing	Electric field, $ B $, ∇B , 0–25 Hz	G. Paschmann	MPE Garching
3DA	Electrostatic analyzer	Electrons/ions, 10 eV–25 keV	G. K. Parks	UoW Seattle
EPI	Solid state detector	Electrons/ions, 20–225/400 keV	T. Sanderson	ESTEC Noordwijk
ICI	Time-of-flight spectrometer	Major ions, 15 eV–40 keV	L. M. Kistler	UNH Durham
PCD	Ion emitter	s/c potential control	K. Torkar	IWF Graz
SFD	Scintillating fiber detector	Electrons/ions $> 0.3/6$ MeV	L. Adams	ESTEC Noordwijk

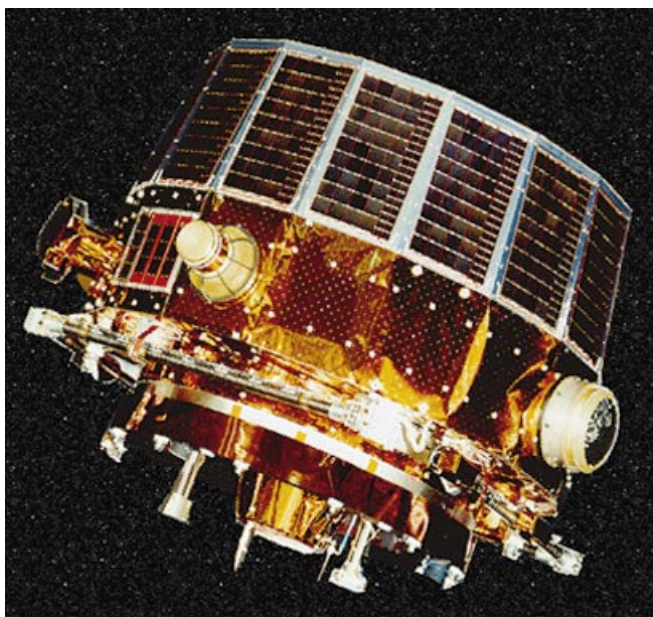


Fig. 1. The Equator-S spacecraft, booms not deployed. One gun-detector unit of the Electron Drift Instrument is seen left from center, to the right the Ion Composition Instrument. Several antennae on the bottom side are extended

similar problem with the ion sensor was feared, the experimenters decided to leave it switched off, until the cause of the electron analyzer failure was clearly established. This took longer than expected, and when the new flight software was ready to be unloaded at the end of April, the whole S/C had gone out of service. So, the payload complement was operated without the 3D ion/electron detectors, except for one brief period during commissioning. This rendered also the potential control device useless, because it had to be controlled by the measurements of the electron sensor. Mass-resolved 3D-ion measurements were, however, available from the Ion Composition Instrument.

Again for financial reasons, mission operations were designed so that support by the ground station at Weilheim for data dumps was needed for only 3 hours a day, while commanding was scheduled only every third contact period. The capacity of the mass memory allowed 13.5 hours of data storage in low rate and 3.4 hours in high rate telemetry. Therefore, four types of orbit coverage were designed:

1. low rate data coverage starting at about $5.5 R_E$ and extending beyond apogee,
2. similar to (1) but with a low altitude hour of high rate data taking,
3. same as (1) but starting apogee,
4. symmetrical around apogee with 90 min of high rate data.

From time to time perigee data were taken for attitude determination.

This strategy restrained the Equator-S science to altitudes essentially outside the radiation belts, quite consistent with its payload complement. Figure 2 shows

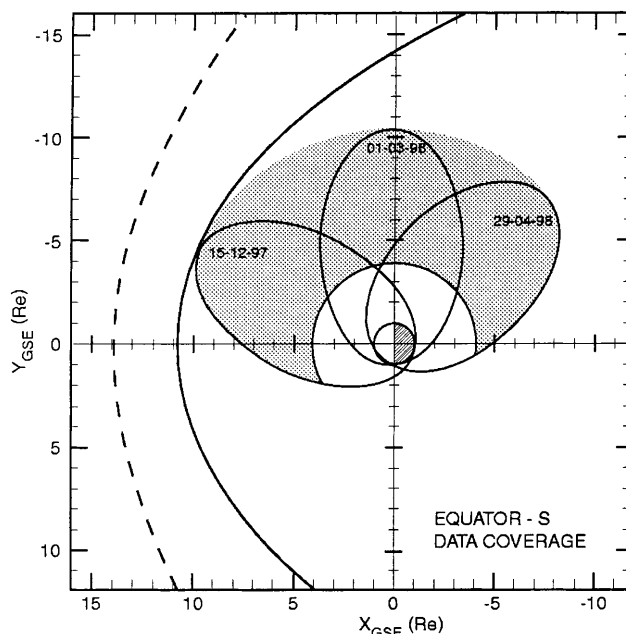


Fig. 2. Equator-S operational orbits projected into the GSE X-Y plane. Grey shading indicates the region of data coverage

in GSE X-Y-coordinates the orbits during the active phase of the spacecraft. The region of data coverage is grey shaded.

Standard processing of the Equator-S data is carried out at the German Cluster Science Data Center at MPE, which has taken on also the task of an Equator-S Data Center (EDC). It obtained the telemetry data received and processed at GSOC directly via a leased ISDN line, and also via CD-ROM. With software provided by the instrument teams, the EDC then generates two sets of key parameters, both closely modelled after the Cluster data sets: the Summary Parameter Data Base (SPDB) with 1-minute time resolution, and the Prime Parameter Data Base (PPDB) with 1.5s (i.e., spin) resolution. The SPDB and PPDB contain ion moments, energetic particle fluxes, and magnetic and electric field values. A standard set of plots of SPDB parameters is generated at the same time. The EDC also archives the data. Once validated by the Lead Investigators, the key parameters can be obtained via NASA's or MPE's CDAWeb. The one-minute data (SPDB) is available to everybody, while access to the 1.5-s data (PPDB) is initially restricted to the Equator-S Science Team.

5 First results

In spite of the short operational life of the Equator-S spacecraft and its reduced operational payload, all three major objectives named in Section 2 have been fulfilled. The EDI technique has been successfully proven in space flight, the data obtained during the five months of operation will make a substantial contribution to the ISTP, albeit restricted to the morning sector of the magnetosphere, and the high time resolution has been successfully exploited. This is documented in the fol-

lowing twelve papers of which we will here give a very brief preview.

The papers by Quinn *et al.* and Paschmann *et al.* describe the two complementary techniques to measure the electric field by (a) triangulation and (b) time-of-flight measurements of the electron beams from two gun-detector units on board the spacecraft. The validity of the techniques in weak and highly variable magnetic and electric fields is demonstrated and data of fields of only a few tenths of mv/m are presented. So $E \times B$ -drifts two or more orders of magnitude smaller than the thermal velocity of the ions can be precisely measured. The time-of-flight technique, in addition, delivers values of the magnitude of B with an accuracy of better than 1% and can thus be applied to determine magnetometer offsets along the spin axis. The amount of valid electric field data obtained during the lifetime of Equator-S is, however, small, because the learning how to operate these complex instruments with their closed-loop control algorithms was quite time consuming. Furthermore, stray fields set up by the operation of the magnetometers, which was needed during the initial 5 months, compromised the exact pointing of the electron beams in the weak fields of the outer magnetosphere. In total, invaluable experience was gained for the operation of the electron drift instruments on the Cluster II mission.

Important as baseline instruments and necessary providers of real-time field orientations for the Electron Drift Instrument were the flux-gate magnetometers (Fornacon *et al.*). They combined a wide dynamic range with high accuracy at frequencies up to 64 Hz and high stability. These excellent properties allowed, for the first time, to determine with flux-gate magnetometers the wave forms of so-called lion roars found typically in the troughs of magnetosheath mirror waves (Baumjohann *et al.*). Wavelet analysis of a great number of events showed that these electron whistler mode waves exist preferentially in the frequency range of 0.05–0.15 f_{ce} (f_{ce} = electron gyrofrequency) and propagate almost precisely along the ambient field direction. In the high-beta environment in which they are found to occur, only small temperature anisotropies suffice for their excitation.

The other contributions in this issue relate to the ISTP. The long dwelling time near the magnetopause gave rise to careful determinations of the position and dynamics of the magnetopause during morning hours (Dunlop *et al.*) and to a thorough study of the occurrence and characteristics of mirror modes outside, but close to the magnetopause (Lucek *et al.*). It is found that the location of the magnetopause scales very well with ram pressure and has mostly rather stable normal vectors. The mirror modes, large amplitude compressional structures, are found to be parallel to the magnetopause. Because of lack of plasma measurements in the initial phase, these studies were based solely on the magnetometer data. Fortunately, before the above-described failure of the 3D electron analyzer happened,

valuable measurements of the 3D plasma analyzer and the solid-state telescopes could be performed during multiple crossings of the magnetopause (Parks *et al.*). The accurate measurement of electrons from 7 to 100 eV was enabled by the operation of the potential control device reducing the spacecraft potential to near zero (Torkar *et al.*), when normally the energy spectra are contaminated by photoelectrons.

Later in the mission, when Equator-S had its apogee in the morningside plasma sheet, field-aligned very high-beta structures, named plasma blobs, were observed which appear to be the magnetospheric relatives of the well-known mirror modes in the magnetosheath (Haerendel *et al.*). The elucidation of the large scale morphology and motion of these blobs was made possible by simultaneous measurements of the Geotail spacecraft during close conjunctions with Equator-S. Such a conjunction during a substorm extending into the late night sector also allowed to analyze the reaction of the plasma sheet at this late time, in particular the flow reversals upon transits from plasma sheet to the lobe (polar cap) field lines (Nakamura *et al.*).

Equator-S was instrumented mainly with the goal to study the outer parts and boundary regions of the magnetosphere including the ring current. In the latter region ion composition measurements were made in the recovery phase of a magnetic storm. Changes in the energy spectra of H^+ and O^- were interpreted in the light of charge exchange losses along their energy-dependent drift paths (Kistler *et al.*). This led to a critical comparison of existing electric field models. The last paper in this series describes the findings with a new technique, a scintillating fibre detector, designed to measure integral fluxes of more than 400 keV (Cyamukungu *et al.*). The whole operation period of Equator-S is covered, and the correlation of the energetic electron belt population with geomagnetic storms is demonstrated.

In summary, Equator-S covers a wide range of aspects, new measuring techniques, detailed studies of the magnetopause, and adjacent magnetosheath, hitherto unknown structures and dynamics of the morningside plasma sheet, and investigations of ring current and radiation belts. In view of these initial results (more are to come), one can only deplore the short life of the spacecraft.

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References

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